

# Wing Morphing Senior Design Team

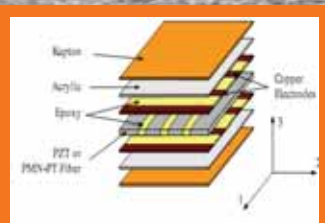
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## SMART MATERIALS

- > Macro Fiber Composites (MFCs) were used due to their relatively high actuation and durability.
- > The MFCs were arranged in a bimorph configuration to maximize deflection.



MFC Bimorph

## MARKET RESEARCH

- > Aircraft was selected to appeal to the majority of modelers
- > R/C forums and magazines were scrutinized to determine most desirable aircraft characteristics (maneuverability, speed, fly anywhere)
- > Electric propulsion system means no exhaust residue



Great Planes Edge 540

## ELECTRONICS

- > MFCs operate between -500 to 1500 volts DC.
- > RC receivers and servos operate on a PWM (Pulse Width Modulated) signal with an amplitude of 0.7 Volts.
- > A printed circuit board (PCB) capable of converting PWM-DC was designed and constructed



PCB Board

## MISSION

**Problem:** Mechanical control systems on small unmanned aircraft are complex and lack resiliency.

**Goal:** Demonstrate the abilities of morphing materials technology by modifying an R/C aircraft through design and fabrication of control surfaces that use Smart Materials.

## WIND TUNNEL TESTING

- > The test was conducted in the Virginia Tech Stability Wind Tunnel on April 15, 2010.
- > The goal of this test was to determine roll rates for the baseline aircraft (the Edge 540) and the morphing wing aircraft.
- > Results show that the baseline configuration was able to achieve a 40% higher roll rate than the morphing configuration at full deflection. This is mostly due to the limited deflection of the MFCs.



Morphing Aircraft at Takeoff



Model Mounted in Wind Tunnel

## FLIGHT TESTING

- > A series of flight tests were conducted at VT Kentland Farms
- > Baseline, partial morphing, and full morphing configurations were successfully flown

## DESIGN

- > Airfoils of varying thickness were generated using Matlab
- > Aerodynamic characteristics of the morphing airfoils were determined using a panel method (XFOIL, Drela, MIT)
- > Stability derivatives were found using a vortex lattice method (AVL, Drela and Youngren, MIT)



AVL Stability Model



CAD Model

## FABRICATION

- > A SolidWorks CAD model was developed
- > Wing and stabilizer components were laser cut
- > Lifting surface structure was fabricated from balsa wood, plywood, basswood, and fiberglass
- > MFCs were bonded with epoxy to a steel substrate to form the control surfaces



Morphing Rudder



Morphing Wing Fabrication